

Appln. of: Friedl et al
Serial No.: 10/828,502
Filed: April 21, 2004

AMENDMENTS TO THE CLAIMS

1. (Currently Amended) A high-pressure turbine of a gas-turbine engine comprising:

a turbine disk carrying rotor blades and rotor blade platforms,

a stator ring carrying stator blades and stator blade platforms,

a lateral wheel cavity formed between the turbine disk and the stator ring, and

a seal provided in an axial direction between the stator blade platforms and the rotor blade platforms which is arranged radially outwardly from a center axis of the high-pressure turbine and adjacent a main gas duct;

wherein the lateral wheel cavity is a single cavity, the rotor blade platforms form a seal runner and the seal is a crocodile-type segmented labyrinth seal with labyrinth tips positioned on the blade platforms.

2. (Cancelled)

3. (Cancelled)

4. (Cancelled)

5. A high-pressure turbine in accordance with Claim 12, wherein the ~~seal is a labyrinth seal, with labyrinth tips attached to the stator ring and with the rotor blade platforms~~ forming a segmented seal runner.

6. (Cancelled)

7. (Cancelled)

8. (Cancelled)

Appln. of: Friedl et al
Serial No.: 10/828,502
Filed: April 21, 2004

9. (Cancelled)

10. (Cancelled)

11. (Currently Amended) A high-pressure turbine in accordance with Claim ~~12~~, wherein the seal includes ~~is of a brush type, with~~ brush elements attached to the stator ring and with the rotor blade platforms forming a segmented seal runner.

12. (Currently Amended) A high-pressure turbine of a gas-turbine engine comprising:

a turbine disk carrying rotor blades and rotor blade platforms,

a stator ring carrying stator blades and stator blade platforms,

a lateral wheel cavity formed between the turbine disk and the stator ring, and

a seal provided in an axial direction between the stator blade platforms and the rotor blade platforms which is arranged radially outwardly from a center axis of the high-pressure turbine and adjacent a main gas duct~~A high-pressure turbine in accordance with Claim 1, wherein the seal is of a brush type, with individual brush elements positioned on the rotor blade platforms.~~

13. (Currently Amended) A sealing arrangement for a high-pressure turbine of a gas-turbine engine having

a turbine disk carrying rotor blades and rotor blade platforms,

a stator ring carrying stator blades and stator blade platforms and

a lateral wheel cavity formed between the turbine disk and the stator ring,

the sealing arrangement comprising a seal provided in an axial direction between the stator blade platforms and the rotor blade platforms which is arranged radially outwardly from a center axis of the high-pressure turbine and adjacent a main

Appln. of: Friedl et al
Serial No.: 10/828,502
Filed: April 21, 2004

gas duct, wherein the seal is of a brush type, with individual brush elements positioned on the rotor blade platforms.